EFFECTS OF HOLD TIME, ANGULAR VELOCITY, PITCH, AND LUMINANCE ON SIMULATED AIRCRAFT IDENTIFICATION RANGE

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ABSTRACT

This paper summarizes the findings from the first of two human factors evaluations conducted as part of the Immersive Display Evaluation and Assessment Study (IDEAS) program. In this evaluation experienced USAF F-16 pilots discriminated and positively identified distant fighter-sized aircraft. On each trial the ownship rapidly approached a pair of aircraft, one "friend" and one "foe," and the observers designated the foe as quickly and accurately as they could.

This evaluation focused on the effects of three variables expected to be primary determinants of motion-induced blurring; hold time, angular velocity of the image, and pixel pitch. An external motion blur reduction shutter was used to systematically manipulate the hold time variable, thus, the luminance of the display system co-varied with hold time. Thus, a fourth independent variable, luminance, was included in the evaluation so that it's effect could be separated from the hold time variable.

Prior to conducting the evaluation a computational model was prepared and used to make quantitative predictions of the effects of these design variables. The correlation between the model predictions and the results of the first evaluation was high (e.g., $R^2 > 0.75$, $p < 0.001$, 109 df). After tuning three parameters in the model to the data the correlation increased significantly ($R^2 = 0.973$, $p < 0.001$, 106 df).

A significant benefit provided by the model is the quantification of the interactions among the design variables. Thus, the model is useful for examining the impact of design trades among the variables that affect task performance.

A summary of this evaluation was published at the IMAGE 2011 conference. This report contains more of the details of the evaluation, the instructions to the observers, and a table of the mean data collected for the 220 experimental conditions.

INTRODUCTION

This paper addresses the effects of five practical display design variables on the range at which pilots can identify aircraft: a visual task of great importance in the training of Air Force pilots. Few would argue that target identification range is not dependent on display resolution. The 5 m minimum dimension of a fighter sized aircraft viewed at a range of 3 km (2 nm) subtends an angle of 5.7 arcmin. At the typical resolution of training display systems of the past decade (e.g., pixel pitch = 2.5 arcmin) the minimum dimension of the aircraft would be 2.3 pixels, far less than the 13-ish pixels recommended by Johnson[8] for target identification tasks.

The use of Johnson's criteria assumes the threshold visual angle for target identification scales linearly with system resolution. Since Johnson's original paper, many similar studies have confirmed the utility of this simple method of analysis[5]. However, it has been pointed out that resolution requirements produced by the method are not precise as they depend on additional factors such as stimulus duration, background clutter, and observer capability[3].

Two recent works have confirmed the linear scaling assumption for the case of relatively coarse pixel pitch where performance is limited primarily by display resolution. However, as pixel pitch is reduced performance becomes primarily limited by observer capability as illustrated in Figure 1. The upper curve in the figure shows the data from of Gaska et. al.[6] for a triangle orientation discrimination task. The lower curve shows the results of one of our preliminary evaluations for a Landolt C orientation discrimination task.

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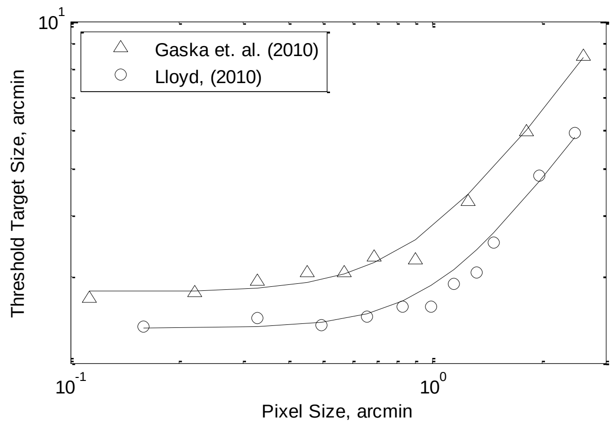


Figure 1. Threshold target size as a function of pixel pitch for triangle and Landolt C orientation discrimination tasks. Threshold target size is proportional to pixel pitch for pitches larger than about 1 arcmin. Threshold target size is constant for pitches below about 0.6 arcmin where it is mediated by observer acuity.

For pixel pitches larger than about 1 arcmin, these data show threshold target size is proportional to pixel pitch. In other words, the observer needs some minimum number of pixels across the target to accomplish the task. For pixel pitches below about 0.6 arcmin, pitch has no effect as performance is limited not by the display system but by the capability of the observer. For the case of static images, the expected effects of pixel pitch are by now well studied. In contrast, very little data are available which indicate how visual performance is affected by pixel pitch in the presence of image motion.

Motion Induced Blur

Motion induced blur has been recognized as a significant limitation of the "sample and hold" projectors (e.g., LCD, LCoS, and DLP) which are now being installed in many simulation trainers. Motion induced blurring occurs when an observer visually tracks a moving target that is drawn using pixels that remain on for a significant fraction of the frame time. Much research pertaining to the causes and remedies for motion induced blurring has been completed by researchers supporting the entertainment and advertising industries. Several recent papers provide overviews of the motion picture response time (MPRT) and related metrics and available methods for measuring the data required for computing them[2, 4, 14, 15, 16].

The International Committee for Display Metrology is expected to release their Display Measurement Standard[23] in the summer of 2011. This standard addresses the MPRT and related measures as well as several methods for their measurement. Concurrently with the development of these methods, the Air Force Research Laboratory (AFRL) in Mesa has conducted a series of evaluations that have focused on correlating a similar metric (hold time) with perceived blur and task performance[5].

Our preliminary evaluation of the standard indicates the measurement procedure should be no more complex than the AFRL-developed procedure. A strong correlation between the MPRT and hold time metric is anticipated as the MPRT is a measure of hold time convolved with the temporal step response of a display. In a future paper, we plan to address the relationship between MPRT and hold time more rigorously and expect to develop a conversion between the two methods of characterizing motion induced blurring so these literatures can be compared.

Pixel hold time

Hold time refers to the duration of time a pixel (and illumination system) is turned on at the commanded state during each frame period. A decade ago researchers at the AFRL developed a simple procedure for measuring hold time in which a fast photo sensor is used to measure a small portion of the screen. The luminance response of the display system is measured for a test pattern that alternates between full on and full off every other frame. The hold time is simply the width of the "on" time of the display device where width is defined using 50% peak luminance points on the measured curve. In the language used by the broader display community, the periodic temporal impulse response (TIR) of the display system is measured using a stationary pattern and stationary sensor. Hold time is computed as the half maximum width of the measured impulse response.

Correlation of hold time and perceived blur

A number of authors have demonstrated a strong correlation between MPRT and perceived blur[4]. Similarly, the data from a series of six evaluations at the AFRL demonstrate the strong relationship between hold time and perceived blur (Figure 2) as measured using a 2-line test pattern for which observers adjusted the width of the gap between the lines[5]. The AFRL evaluations indicate this relationship holds over a range of display technologies including CRT, LCoS, and DLP projectors.

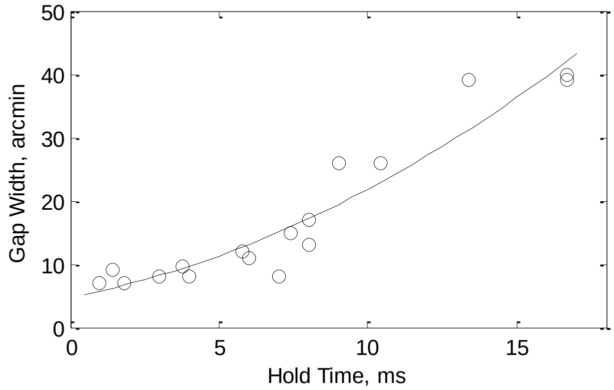


Figure 2 Gap Width measured using the 2-line perceptual blur test, as a function of Hold Time, showing a correlation of $R^2 = 0.91$ ($p < 0.001$, 16 df). Data are from six separate evaluations as summarized in Figure 8.3 of Gaska, et. al (2010) for a line speed of 40 deg/sec.

Correlation of hold time and task performance

While we expect hold time to correlate well with training task performance, relatively little work has been done to demonstrate this correlation. In a study conducted by Winterbottom et. al.,[18] aircraft roll detection threshold was measured as a function of hold time. The correlation obtained in this evaluation was moderate but statistically reliable ($R^2 = 0.4$, $p = 0.03$, 10 df). To date we have found no other papers describing evaluations in which task performance was measured as a function of hold time or MPRT. Thus, we do not yet have sufficient data to recommend the use of the hold time metric (or the MPRT) for the evaluation of simulation training display systems on the basis of task performance.

Model of Task Performance

In early 2010, work restarted on the development of a computational model of visual performance for display systems. This model is an extension of decades of image quality metric development work by Snyder, Barten, and their colleagues during the 80s and 90s[1, 11, 12, 13]. An overview of this model is provided in a companion paper at this conference[10] and more detailed descriptions of the validation studies summarized in this paper will be provided in technical reports in preparation[20, 21].

At the heart of the task performance model is the calculation of the limiting resolution of the display system. A primary input to this calculation is the modulation transfer function (MTF) of the display system which is typically computed from a measured line spread function (LSF) of the display (See Figure 5 for example). Other inputs include angular pixel pitch, hold time, target velocity, contrast, luminance, noise, and anti-aliasing. The parameters MTF, pixel pitch, hold time, target velocity, contrast, and anti-aliasing are used to compute the system MTF. The parameters luminance and noise are used to compute the contrast threshold function (CTF) of the observer. The crossover point of the system MTF and CTF is used to determine the limiting resolution of the display system which is used in the calculation of identification range.

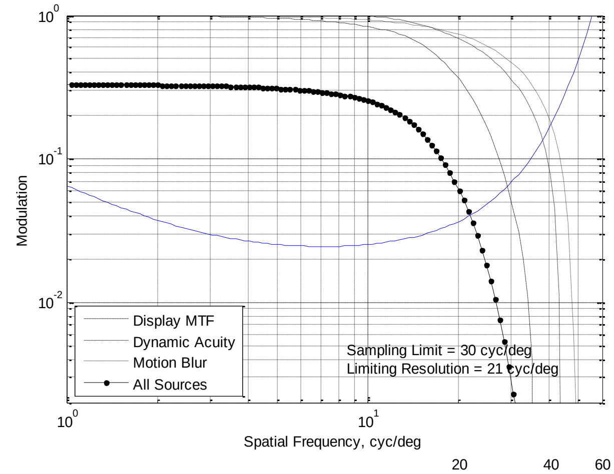


Figure 3. Illustration of the essential calculations performed within the task performance model. In this example the limiting resolution of the system is 21 cyc/deg for a pixel pitch of 1 arcmin, hold time of 8 ms, velocity of 5 deg/sec, luminance of 10 fL, and target CR of 2.2:1.

METHOD.

Participants

A total of eight male observers participated in the first evaluation. Seven of the eight observers were experienced USAF F-16 pilots. The observers ranged in age from 31 to 48 years with a mean age of 43 years. Prior to participation, the (far) visual acuity of each observer was measured using an Optec 2000 vision tester. All observers had a visual acuity of 20:17 or better. The single non-pilot observer was well practiced with the identification of the aircraft models and had an acuity of 20:12.

Evaluation Task

In this evaluation, a self-paced, two-alternative, forced choice procedure was used in which the participant selected the “foe” on each trial as quickly and accurately as practical. On each trial, the ownship started at a range of 3 to 6 km from a pair of aircraft, one friend and one foe. The starting position for each trial was set at 2.2 times the expected identification range and was randomized +/- 20 percent. Trial length was capped at a maximum of 15 seconds. On average the participants identified the foe after about 7-8 seconds and initiated the next trial immediately. Each observer participated in two experimental sessions on separate days.

On each trial, the aircraft traveled in straight and level flight at a ground speed of 250 to 300 knots, pointing either left or right as in Figure 4. The mean contrast of the aircraft against their background was 2.2:1.

At the typical identification range, the horizontal speed of the bogies would produce a negligible angular velocity from the point of view of the ownship.

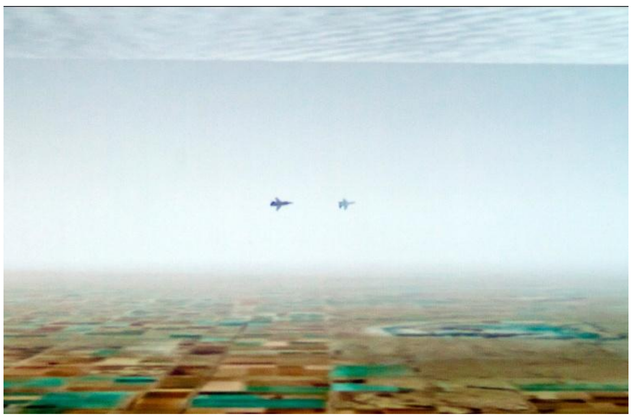


Figure 4. Photograph of a typical trial showing a pair of aircraft pointing to the right. The aircraft were always seen against the uniform portion of the sky.

The angular velocity of the targets/image was set to a constant and controlled level on each trial by changing the pitch and yaw of the ownship in a circular orbit. From the point of view of the observers, this gave the appearance of the ownship approaching the bogies in a spiraling motion. This orbital motion of the ownship allowed sustained high angular velocities for the duration of the trial while keeping the targets near the center of the screen. A second advantage of the spiraling motion was that it produced motion smearing in all orientations during the course of each trial. Angular velocity was controlled by the diameter of the orbit. The largest orbit used in the evaluations had a radius of 8 inches which kept the bogies within the central portion of the screen where our calibration of the hold time shutter was valid.

Prior to each experimental session, each observer studied larger images of the aircraft to become familiar with their appearance. Each session required about 50 minutes to complete.

Equipment

Image generator

The IG computer ran the Windows XP operating system on a custom built computer, consisting of an Intel Core I7 - 920 processor 12 GB of ram. The graphics for the IG were driven by the Nvidia Quadroplex 2200 D2 model which provided the 4 channels required to drive the Sony SXRD projector at a resolution of 4096 X 2160. The IG software is MetaVR version 5.6. The Sim Host computer ran the MATLAB (The Math Works) software under the Windows XP operating system.

Projector and Screen

Both evaluations were conducted in the OBVA laboratory at the AFRL facility in Mesa AZ using an 8 Mpix Sony SRX-S110 LCoS projector. The image was projected on a flat screen measuring 2.28 x 1.27 m (90 x 50 in). The center of the screen was 1.88 m (74 in) above the floor. The projector was mounted overhead on a stand which position the lens 2.39 m (94 in) from the floor and 6.3 m (248 in) from the screen. The walls in the laboratory were painted black, thus, very little scattered light was present.

Motion blur reduction shutter

An LCD motion blur reduction shutter was purchased from VDC Display Systems in the fall of 2010. This device allows hold time to be controlled, from trial to trial, over a range of 1.5 to 14 ms.

Five levels of hold time, indicated in Column 1 of Table 1 were used in the first evaluation. The luminance of the projected image varied proportionally with the hold time setting of this shutter as expected. The luminance levels for the sky against which the aircraft were observed are indicated in Column 2 of Table 1.

With the Sony projector used in this evaluation the image is drawn from the horizontal centerline out. In other words, at the beginning of each frame the image begins updating along the central horizontal line separating the four quadrants of the display system. This method of updating the image must be taken into account when characterizing the effect of the hold time shutter. Since the hold time shutter does not follow the same spatio-temporal pattern of image update that the projector uses, use of the shutter produces different hold times for different vertical positions in the image.

For short hold time settings of the shutter it is possible to achieve the same hold time over most of the vertical extent of the image. As hold time increases, the vertical extent of the image over which a constant width, uni-modal pulse of light is created decreases.

Hold time was measured using a pair of silicon photodiodes with a very fast response time

One photodiode was vertically positioned at the center of the screen while the second was lowered (or raised) from the centerline.

In this evaluation our goal was to exercise hold time over a wide range. We found that could obtain uni-modal temporal waveforms if we used a maximum hold time of 12 ms and restricted our use of the screen to +/- 8 inches from the horizontal centerline. With these constraints applied, we measured the temporal responses as shown in Figures X to X.

For these measurements the image generator was set to produce a full screen white image alternating with black every other frame. The plots below show the projector/shutter response over a period of 60 ms (3.6 frames). The upper trace in each figure shows the response at the horizontal centerline of the screen and the lower trace shows the response for the sensor placed 8 inches below the centerline of the screen. Hold time is defined as the half-maximum width of these functions.

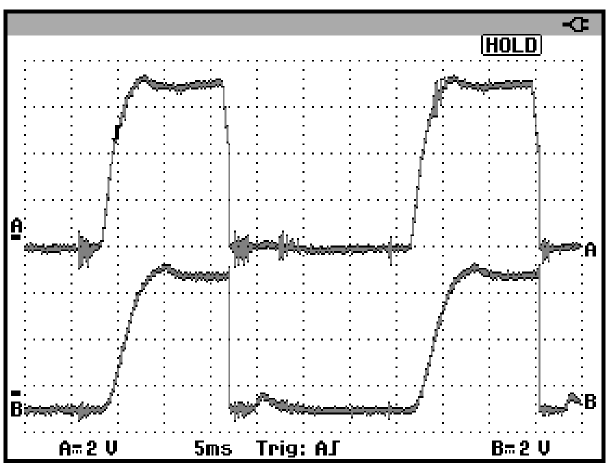


Figure X. Luminance over a period of 60 ms for a long hold time condition (nominally 12 ms). Note that the hold time at the center of the screen was about 10% longer than the hold time at a position 8 inches down from the centerline of the screen.

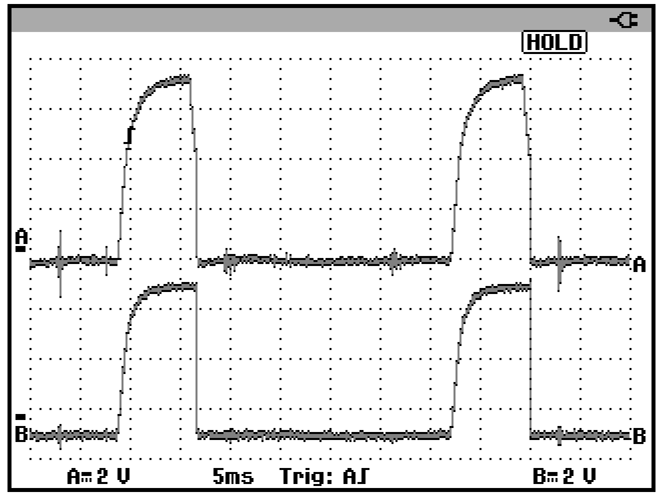


Figure X. Luminance over a period of 60 ms for an intermediate hold time condition (nominally 7.5 ms).

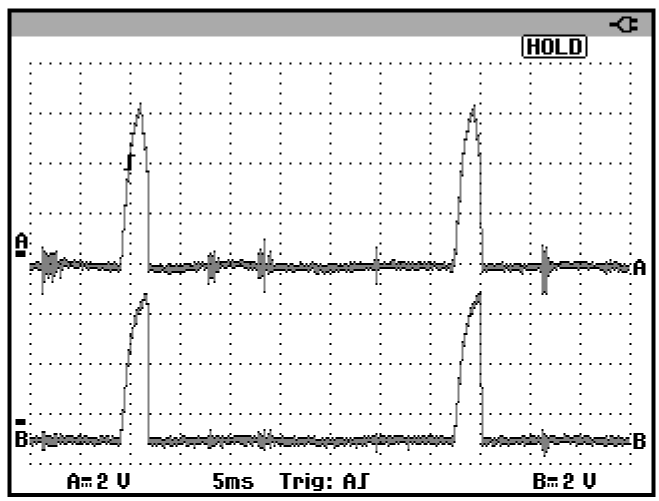


Figure X. Luminance over a period of 60 ms for a short hold time condition (nominally 2 ms).

The hold time variable was measured and calibrated by fixing the “projector delay” at 100 and varying the “turn on delay” over a range of values from 250 to 1400. Hold times were measured ranging from 12 to 2 ms and were related to the projector delay setting of the shutter by fitting a regression line. The measured levels of hold time were found to be accurately linearly related to the projector delay settings and the resulting equation was used to compute the shutter setting for each hold time condition:

Turn on delay = 1622 – 116 * hold time

Filters and Luminance

In the first evaluation display luminance was confounded with the hold time setting of the shutter. Since there would be no way to differentiate the effects of luminance and hold time, luminance was also independently manipulated. This was accomplished by doubling the number of conditions in the first evaluation and reducing luminance to half for the additional trials using a neutral density (ND) filter mirror. The resulting luminance levels for the filtered conditions are provided in Column 3 of Table 1.

Table 1. Mean luminance levels of the sky against which the bogies were observed for all combinations of the hold time and filter condition. These levels were approximately 85% of the peak white of the display system.

Hold Time, ms	Luminance with no Filter, cd/m^2	Luminance with Filter, cd/m^2
2.0	11	5.5
4.5	24	13
7.0	45	22
9.5	58	29
12.0	76	39

Pixel Pitch and Resolution

In this evaluation, the practical variable viewing distance was used to control the angular pixel pitch of the display system. No other manipulations were made that independently affected the relationship between pixel pitch, viewing distance, and measured resolution (MTF), thus, the three variables Pixel Pitch, Viewing Distance, and Resolution were completely confounded and are used inter-changeably in this report.

For most training display systems, the linear Pixel Pitch (in mm) and viewing distance are clearly defined and relatively immutable attributes of the system. The angular Pixel Pitch (in arcmin) of the system is easily computed from these two quantities and is thus also clearly defined and not often misinterpreted.

However, the "effective" or "limiting" resolution of a display system is not nearly as easy to define or measure as is pixel pitch. This is primarily because this system attribute depends on a number of additional factors such as optical blur, pixel hold time, angular velocity, mis-convergence, luminance, contrast, anti-aliasing, and observer acuity.

For these evaluations, the pixel pitch (pixel-to-pixel spacing) measured at the center of the screen was 0.60 mm (100 pixels measured 60 mm). The vertical and horizontal pitches differed by no more than 2%.

For each trial, the observer was seated at one of five viewing distances indicated in Column 1 of Table 5. The angular pixel pitch of the display system corresponding to each viewing distance is provided in Column 2. To give the reader a sense for the number of pixels used to render images in this evaluation, Column 3 provides the number of pixels spanning the wingspan of the aircraft.

Table 5. Viewing distance, angular pixel pitch and number of pixels per wingspan for a fighter aircraft with an 11 m wingspan viewed at a range of 2.5 km (15 arcmin).

Observer – Screen Distance meters	Pixel Pitch arcmin	Number of Pixels per 15 arcmin wingspan
0.8	2.58	5.8
1.2	1.72	8.7
1.8	1.15	13
2.8	0.74	20
4.2	0.49	30

The line spread function (LSF) of the projected image was measured using a calibrated color camera (Canon G-9) positioned approximately 12 inches from the screen (see Figure 5). For this measurement, a pair of widely spaced single pixel wide white lines on a black background were projected on the screen and photographed. The space between the pair of lines was measured with a ruler and used to determine the sampling rate of the camera arrangement which measured 11.4 camera pixels per mm.

The MTF of the display system was used as an input to the model along with the settings of each of the five independent variables used in the evaluations. The limiting resolution of the display system and the expected threshold target size were computed separately for each of the 420 experimental conditions.

Aircraft Models

Table 6 lists the model names for the 11 friendly and 10 enemy aircraft used in the first evaluation. On each trial one aircraft was selected at random from each of these lists. The aircraft model numbers were recorded for each trial so that a table of the relative discriminability of model pairings could be constructed. An average of 31 trials were used to estimate the level of each aircraft pairing and these data were used to remove the variance due to the pairings.

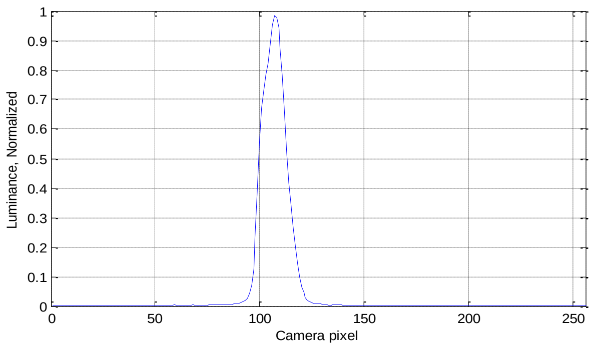


Figure 5. Line spread function for a single white line on a black background. Camera calibration was 0.088 mm / camera pixel, thus, about 25 camera pixels spanned the line spread function.

Table 6. Listing of the 11 friendly and 10 enemy aircraft models used in the evaluation. All models were supplied with the MetaVR image generator.

Friendly Aircraft		Enemy Aircraft	
120	F-16	131	Mig 29
121	F-16_Oman	132	Mig 29C
122	F-2	133	Su 30
123	F-18	134	Su 30mk
124	CF-18	135	Su 27
125	F-16_Low_res	136	Mig 25
126	F-35	137	Mirage_Iraq
127	F-15	138	Mig 23
128	AV8B	139	Mig 21
129	Tornado_F3	140	Mig 21_Iraq
130	Tornado_gr1		

Independent Variables

The primary goal for the experimental design was to cover the design trade space, making sure to gather enough data to fully characterize the expected interactions among the variables. The independent variables and levels used in the first evaluation were:

- Hold Time, 5 levels: 2, 4, 7, 9.5, and 12 ms
- Target Velocity, 5 levels: 1, 8, 15, 22, and 29 deg/sec
- Pixel Pitch: 2.57, 1.70, 1.14, 0.73, 0.49 arcmin
- Luminance (filter), 2 levels, 100 and 50%

A nearly full factorial design was used so that variance representing the interactions would be included in the data. 220 of the 250 possible combinations were measured; the remaining 30 combinations were not collected because the required orbit diameters were larger than the +/- 8 inch central portion of the screen where the blur reduction shutter calibration was considered valid.

RESULTS

At all stages of analyses, the distributions of data were clearly more symmetrical and Gaussian when a logarithmic transformation of threshold size was used as compared with either size or range. Thus, all of the statistical analyses described below were performed on the log10 transformation of the dependent variable to improve the accuracy of the statistical tests.

Prior to conducting the statistical analyses reported below, the effects of observer, practice, and aircraft model pairings, were removed from the data. Details of the data reduction procedures used for each evaluation are provided in the AFRL technical reports describing each evaluation[20, 21].

For the first evaluation, the data from 3511 trials were used in the analyses, for an average of 16 trials per experimental condition.

Fit of Initial Model

Prior to completing the data collection for this evaluation the model described in Lloyd et. al., (2011)[10] was used to compute the expected responses to the experimental variables. Prior to the optimization of model coefficients, the correlation between these predictions and the mean responses of the 8 observers was $R^2 = 0.78$ ($p < 0.001$, 109 df). In the analyses that follow, the model coefficients were optimized to maximize the correlation between the model and data.

Differences among observers

The eight observers in the first evaluation differed substantially in the maximum ranges at which they identified targets. The mean range for the three observers with the longest ranges was 1.6 times larger than the mean range of the three observers with the shortest ranges. The correlation between identification range and visual acuity was in the expected direction; however, it was not statistically reliable ($p > 0.05$, 7 df). Thus, it appears that factors in addition to acuity may be responsible for the differences among observers.

Effect of luminance

The independent variable luminance (filter) had a small but statistically reliable ($p < 0.01$, 109 df) effect on performance. When luminance was halved with the insertion of the ND filter, the mean size at which targets were identified increased by 2.5%. Halving the luminance of the display system using the filter essentially raised the surfaces plotted in Figures 6 and 7 by 0.011 log10 units without noticeably changing their shape.

Effects of Hold Time, Angular Velocity, and Pixel Pitch

Compared with the effect of Luminance, the variables Hold Time, Angular Velocity, and Pixel Pitch had much larger effects as predicted by the model. Since the interactions among these variables are strong and complex, the effects of all three variables are shown in the form of surface plots which illustrate the main effects and interactions in the same set of plots. For these plots the data have been averaged over the two levels of luminance which did not substantially change the shapes of the surfaces.

The results of the evaluation are illustrated in Figures 6 and 7 which show the effects of pixel pitch and hold time on the log10 of threshold identification size for two angular velocities. These plots illustrate the fit of the model (the surfaces) to the data. The circles on these figures indicate the mean threshold target size averaged across observer and filter condition (N = 16). The correlation between these data and the model was R^2 = 0.973 ($p < 0.001$, 106 df). The standard deviation of the residuals was 0.021. Converting from log10 of the residuals, the standard deviation was 4.9% of the target size (or range).

When the data were averaged across observer and not filter condition (N = 8), the correlation between data and model was R^2 = 0.952 ($p < 0.001$, 216 df). For this case the standard deviation of the residuals was 0.028. Converting from log10 of the residuals, the standard deviation was 6.6% of the target size (or range).

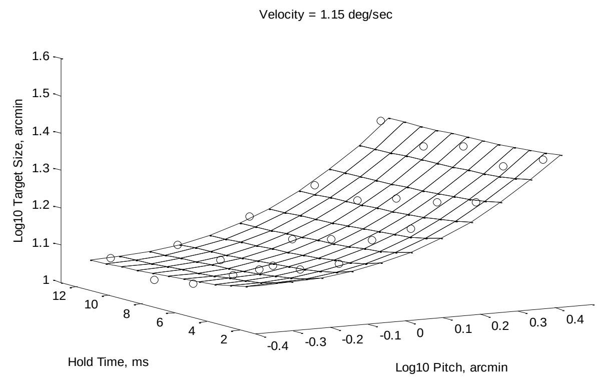


Figure 6. Threshold target size as a function of Pixel Pitch and Hold time for a velocity of 1 deg/sec. Note that hold time had little effect for slowly moving targets.

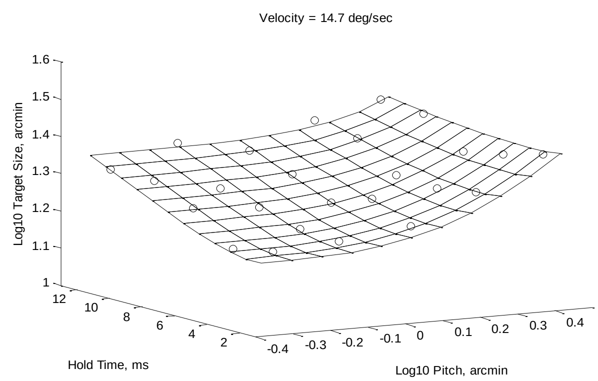


Figure 7. Threshold target size as a function of Pixel Pitch and Hold time for a velocity of 15 deg/sec. Note that at 15 deg/sec hold time has a large effect on threshold target size, especially for fine display pitch.

DISCUSSION

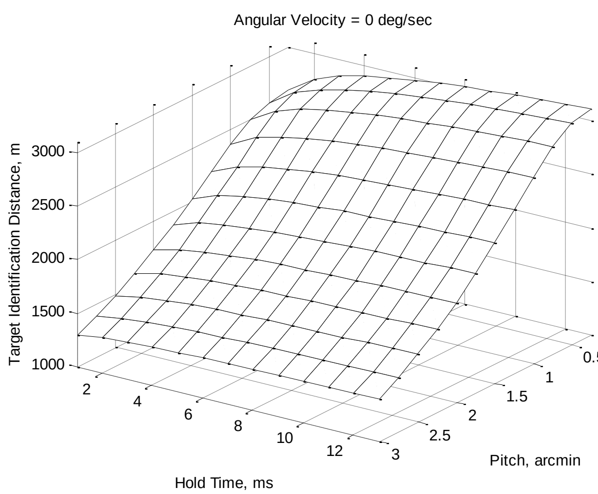
In the previous sections, the data and model were shown together on surface plots that were scaled in a transform space that homogenizes the variance across the experimental conditions so that the fit of the model to the data could be assessed. In this section, we plot several views of the model in a transform space that is more immediately useful to a display system specifier or design engineer.

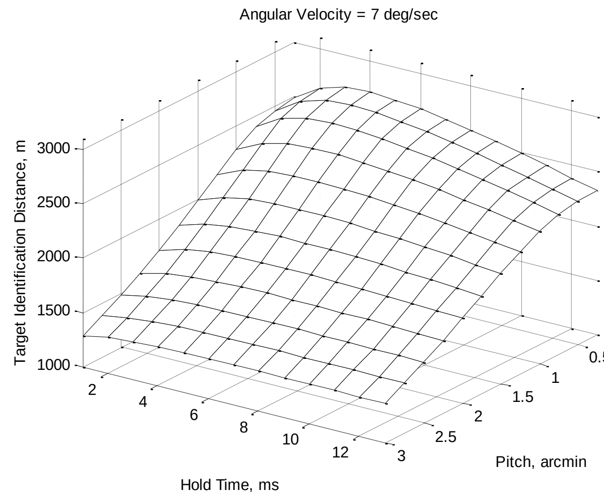
Figure 10 illustrates the effects of the three variables that had the largest impact on performance: Pitch, Hold time, and angular Velocity. These surfaces represent the mean performance of our 8 observers for a peak display luminance of 30 fL, a display contrast of 20, and a fighter-sized aircraft (11 m wingspan).

Figures 11 and 12 illustrate the effects of display luminance and contrast which had smaller effects on performance than did the first three variables. Figure 11 represents the case of dark targets against a bright background that is near the peak luminance of the display system. Figure 12 shows the effect of contrast is expected to be stronger when the target background is only 25% of the peak display luminance.

CONCLUSIONS & RECOMMENDATIONS

- These evaluations provide the simulation training community with far more data pertaining to hold time and task performance than were previously available.
- The expected effect of hold time on task performance has been confirmed with high statistical reliability across hundreds of combinations of parameter settings.
- The correlation between model predictions and the data were very high, confirming the validity of the model.
- The model accurately quantifies the interactions between the five practical design variables, thus, the model is well suited for supporting design trades among these variables.





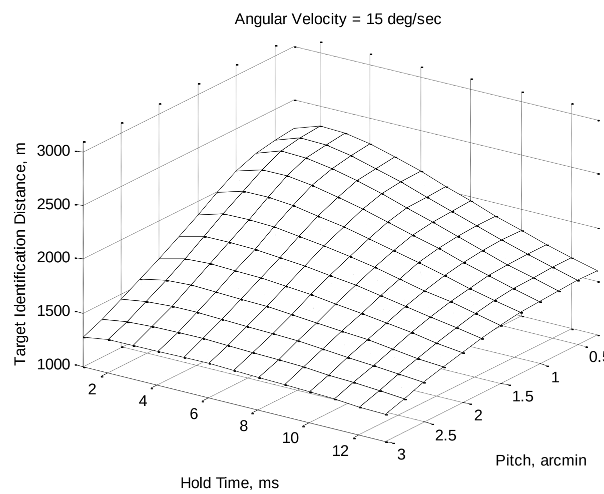


Figure 10. Threshold target identification range for fighter-sized aircraft as a function of Pitch and Hold time for three levels of target Velocity.

Beware of Flicker

To obtain near-eye limited resolution in the presence of even moderate image motion, hold times of only a few ms will be required (see the bottom panel of Figure 10). A substantial literature recommends the use of frame rates of 75 Hz or greater for short hold time displays (e.g., CRTs) to avoid the detrimental effects of flicker[24].

We know of no other means by which motion induced blurring can be reduced to inconsequential levels but to reduce hold time. Thus, it appears the simulation training industry will have to move to higher frame rates as we move towards eye-limited resolution.

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AUTHOR BIOGRAPHIES

Dr. Charles J. Lloyd has 25 years of experience in display systems and applied vision research at such organizations as the Displays and Controls Lab at Virginia Tech, the Advanced Displays Group at Honeywell, Lighting Research Center, Visual Performance Inc., BARCO Projection Systems, and FlightSafety Int. Charles is now the Lead Scientist for the IDEAS program at the Air Force Research Laboratory (L-3 Communications) where he manages the development and validation of display system metrics for simulation training. Charles has presented 65 papers in this arena.

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APPENDIX A: INSTRUCTIONS TO PARTICIPANTS

In this evaluation we are asking you to imagine yourself in the roll of a fighter pilot who has been called to help your wing mate who is embroiled in a battle with an enemy fighter. The "friend" and "foe" have been stuck in a turn circle and have shed velocity down to a few hundred knots. You are approaching their battle space at a much greater velocity from a range of 2 to 4 nm. Unfortunately, you do not know which fighter is the enemy and must determine this visually as soon as possible so that you can take action to assist your wing mate.

Experimenter runs a few trials to demonstrate the procedure

During the evaluation you will be presented with a series of trials in which one bogie is chasing another as you rapidly approach. Your task on each trial is to positively identify the foe as soon as you can and to press the left or right button to indicate the side of the screen that the foe is on. The trial will stop as soon as you make your selection.

Point out the buttons to the participant; let them run a few trials

You can begin the next trial by pressing either of the two buttons. You may take a break between trials as you see fit. During the evaluation we will present about 250 trials which should take us about one hour to complete.

On each trial you will be presented with one friend and one foe aircraft. The aircraft from each category are shown on this chart.

Display the aircraft models and give the participant a minute to study

During each trial we will be systematically manipulating several display design variables so that we can measure their effects on aircraft identification performance. One of these variables is resolution which we control by changing your viewing range. After every 15 to 20 trials we will ask you to move to a different position relative to the screen. These positions are marked on the floor with tape.

Point out the participant seating positions marked on the floor

Another variable we will manipulate during the evaluation is the angular velocity of the bogies, that is, the speed with which the bogies move relative to the projection system. For some of our trials we wish to produce high angular velocities. Unfortunately, we could not afford to set up a very wide and high resolution display system for this evaluation. Our FOV is limited to only a single channel. If we were to use linear motions our bogies would be on and off the screen in less than a second, too short a time for our experimental task.

The only way we know how to produce high angular velocities and keep the targets on the screen for the entire trial is to move the scene in circles. The primary method we use to produce this circular motion through changes in the pitch and heading of the ownship.

We understand that the motions of the ownship are unnatural; in fact, we acknowledge it is probably not possible to get a fixed wing aircraft to move in this way.

Please also accept our apology for not providing you with a set of controls that would allow you to fly the aircraft. Given the limited budget available for this program we were not able to set up a cockpit with controls typical of fast jets.

Assuming you can tolerate the limitations of our experimental setup, we believe the angular velocity effects we measure here will fairly represent the effect of angular velocity when you are in control of the aircraft and can produce more natural movements of the image.

Do you have any questions regarding the procedure?

Answer questions

Prior to starting the evaluation we would like you to complete a series of practice trials so that you can get used to the procedure and can become familiar with the aircraft models. We will conclude the practice trials when you get to the point that you can correctly identify the foe on 15 trials in a row.

Remember that your main goal is to identify the foe as quickly as you can but without making errors.

APPENDIX B: MEAN RESPONSE DATA

The data provided in this table contains the mean data from the 8 participants in the evaluation. Each observer participated in two experimental sessions, thus, there are approximately 16 observations for each mean. Trials on which identification errors were made were eliminated.

Column Descriptions

1. Viewing distance, observer to screen, m
2. Pixel hold time, ms
3. Angular velocity of image, deg/sec
4. Filter condition: 0 = no filter, 1 = filter
5. Log10 of the measured threshold angular subtense, arcmin
6. Standard deviation of the thresholds
7. Number of correct observations for each conditions

Derived Variables

Angular Pixel Pitch

The linear pixel pitch of the projected image remained fixed at 0.60 mm for all conditions used in this evaluation. The angular pixel pitch was manipulated by changing the viewing distance. Viewing distance is converted to angular pixel pitch using the formula:

Pitch = 60 * atand(0.0006 / viewDist)

The five viewing distances used in the evaluation, 0.79, 1.19, 1.78, 2.77, and 4.17 m, produced angular pixel pitches of 2.61, 1.73, 1.16, 0.74, and 0.49 arcmin.

Threshold Identification Distance

During each experimental session the threshold distance was recorded on each trial. Preliminary analyses of the data indicated the data were more normally distributed and the variance distributed more homogeneously across the experimental variables when it was transformed from distance to angular subtense. Thus, threshold distance was converted to angular subtense assuming an 11 m wingspan as the representative dimension of the fighter aircraft. All subsequent analyses were performed on the angular data.

The angular threshold data in Column 5 of Table B can be converted back to threshold distance (meters) using the formula:

threshDist = wingspan ./ tand(10.^logMeasured / 60)

For example, the threshold distances for the first and last conditions in the table are 1678 and 1953 m.

Peak Display Luminance

In this evaluation the peak luminance of the display system varied as a function of two of the independent variables; filter condition and hold time. The peak display luminance (fL) can be calculated using the formula:

peakDislayLum =

35.4 * (1 - 0.49 * filter) * holdTime / 16.67

For example, the peak display luminance levels for the first and last conditions in the table are 4.54 and 25.7 fL.

Table B. Data from First IDEAS Evaluation

0.79	2.14	1.15	0	1.3528	0.0721	13
0.79	2.14	1.15	1	1.4090	0.1335	16
0.79	2.14	7.90	0	1.4061	0.1421	16
0.79	2.14	7.90	1	1.4141	0.1398	16
0.79	2.14	14.70	0	1.3940	0.0931	16
0.79	2.14	14.70	1	1.4180	0.1223	18
0.79	2.14	21.70	0	1.4161	0.0731	16
0.79	2.14	21.70	1	1.4323	0.0854	17
0.79	2.14	28.70	0	1.4672	0.0809	16
0.79	2.14	28.70	1	1.4380	0.1196	15
0.79	4.55	1.15	0	1.3490	0.1207	13
0.79	4.55	1.15	1	1.3259	0.0713	18
0.79	4.55	7.90	0	1.3604	0.1158	15
0.79	4.55	7.90	1	1.3651	0.0915	17
0.79	4.55	14.70	0	1.3677	0.0624	14
0.79	4.55	14.70	1	1.3879	0.1311	15
0.79	4.55	21.70	0	1.4508	0.1004	14
0.79	4.55	21.70	1	1.4475	0.1234	16
0.79	4.55	28.70	0	1.4531	0.0957	13
0.79	4.55	28.70	1	1.4526	0.0823	17
0.79	7.04	1.15	0	1.3480	0.1157	15
0.79	7.04	1.15	1	1.3711	0.1037	17
0.79	7.04	7.90	0	1.3801	0.1314	13
0.79	7.04	7.90	1	1.3769	0.0887	13
0.79	7.04	14.70	0	1.3640	0.0640	15
0.79	7.04	14.70	1	1.3497	0.0923	18
0.79	7.04	21.70	0	1.4381	0.0983	14

0.79	7.04	21.70	1	1.4520	0.1021	16
0.79	7.04	28.70	0	1.4428	0.0779	13
0.79	7.04	28.70	1	1.4724	0.1225	16
0.79	9.47	1.15	0	1.3498	0.1321	12
0.79	9.47	1.15	1	1.3139	0.0936	15
0.79	9.47	7.90	0	1.3329	0.1124	14
0.79	9.47	7.90	1	1.3783	0.1030	15
0.79	9.47	14.70	0	1.4119	0.0781	13
0.79	9.47	14.70	1	1.4481	0.0919	17
0.79	9.47	21.70	0	1.4561	0.0986	14
0.79	9.47	21.70	1	1.4606	0.0953	16
0.79	9.47	28.70	0	1.5312	0.0932	15
0.79	9.47	28.70	1	1.5700	0.1277	16
0.79	12.12	1.15	0	1.3792	0.1029	15
0.79	12.12	1.15	1	1.3597	0.1195	16
0.79	12.12	7.90	0	1.3759	0.0996	15
0.79	12.12	7.90	1	1.3395	0.0840	18
0.79	12.12	14.70	0	1.4351	0.0800	12
0.79	12.12	14.70	1	1.4384	0.0875	16
0.79	12.12	21.70	0	1.5434	0.1262	13
0.79	12.12	21.70	1	1.4736	0.0895	17
0.79	12.12	28.70	0	1.5589	0.0951	11
0.79	12.12	28.70	1	1.5517	0.1193	12
1.19	2.14	1.15	0	1.2691	0.0941	17
1.19	2.14	1.15	1	1.2975	0.0853	19
1.19	2.14	7.90	0	1.2648	0.0606	19
1.19	2.14	7.90	1	1.2739	0.1242	18
1.19	2.14	14.70	0	1.3004	0.0830	18
1.19	2.14	14.70	1	1.3360	0.1015	17
1.19	2.14	21.70	0	1.3597	0.1368	20
1.19	2.14	21.70	1	1.3585	0.1008	18
1.19	2.14	28.70	0	1.3795	0.0913	18
1.19	2.14	28.70	1	1.3698	0.1050	16
1.19	4.55	1.15	0	1.2505	0.0987	19
1.19	4.55	1.15	1	1.2614	0.0889	15
1.19	4.55	7.90	0	1.2460	0.0862	16
1.19	4.55	7.90	1	1.2655	0.1261	16
1.19	4.55	14.70	0	1.3418	0.0951	18
1.19	4.55	14.70	1	1.2606	0.1159	16
1.19	4.55	21.70	0	1.3091	0.1176	17
1.19	4.55	21.70	1	1.3587	0.0894	14
1.19	4.55	28.70	0	1.4337	0.1086	17
1.19	4.55	28.70	1	1.3454	0.0868	15
1.19	7.04	1.15	0	1.2210	0.0874	20
1.19	7.04	1.15	1	1.2542	0.1385	18
1.19	7.04	7.90	0	1.2229	0.0746	18
1.19	7.04	7.90	1	1.2521	0.0853	16
1.19	7.04	14.70	0	1.2861	0.0920	15
1.19	7.04	14.70	1	1.3316	0.0760	16
1.19	7.04	21.70	0	1.3975	0.1045	18
1.19	7.04	21.70	1	1.4091	0.1100	15
1.19	7.04	28.70	0	1.4341	0.0713	16
1.19	7.04	28.70	1	1.4594	0.1169	16
1.19	9.47	1.15	0	1.1975	0.0840	15
1.19	9.47	1.15	1	1.2104	0.1196	14
1.19	9.47	7.90	0	1.2922	0.0854	16
1.19	9.47	7.90	1	1.2418	0.0762	16
1.19	9.47	14.70	0	1.3595	0.0962	16
1.19	9.47	14.70	1	1.3961	0.0784	15
1.19	9.47	21.70	0	1.4489	0.0899	17
1.19	9.47	21.70	1	1.4248	0.1079	16
1.19	9.47	28.70	0	1.4776	0.1375	17
1.19	9.47	28.70	1	1.4777	0.0803	15
1.19	12.12	1.15	0	1.2053	0.0760	16
1.19	12.12	1.15	1	1.2199	0.1177	19
1.19	12.12	7.90	0	1.2449	0.0872	18
1.19	12.12	7.90	1	1.2421	0.0960	14
1.19	12.12	14.70	0	1.4171	0.1302	18
1.19	12.12	14.70	1	1.3769	0.0770	17
1.19	12.12	21.70	0	1.5142	0.1168	18
1.19	12.12	21.70	1	1.4954	0.1225	17
1.19	12.12	28.70	0	1.5655	0.1633	14
1.19	12.12	28.70	1	1.5527	0.1033	15
1.78	2.14	1.15	0	1.1999	0.0850	15
1.78	2.14	1.15	1	1.2548	0.0940	20
1.78	2.14	7.90	0	1.2111	0.0914	16
1.78	2.14	7.90	1	1.2367	0.1058	20
1.78	2.14	14.70	0	1.2452	0.1355	16
1.78	2.14	14.70	1	1.2402	0.0897	20
1.78	2.14	21.70	0	1.2694	0.0963	17
1.78	2.14	21.70	1	1.2890	0.0967	20
1.78	2.14	28.70	0	1.3468	0.0919	15
1.78	2.14	28.70	1	1.3283	0.0978	19
1.78	4.55	1.15	0	1.1253	0.0785	14
1.78	4.55	1.15	1	1.2130	0.1046	17
1.78	4.55	7.90	0	1.1830	0.0979	14
1.78	4.55	7.90	1	1.2267	0.1182	16
1.78	4.55	14.70	0	1.2473	0.1025	15
1.78	4.55	14.70	1	1.3302	0.1086	16
1.78	4.55	21.70	0	1.2777	0.0818	16
1.78	4.55	21.70	1	1.3021	0.0620	18
1.78	4.55	28.70	0	1.3400	0.0938	15
1.78	4.55	28.70	1	1.3645	0.0822	16
1.78	7.04	1.15	0	1.1322	0.0893	14
1.78	7.04	1.15	1	1.1531	0.1018	20
1.78	7.04	7.90	0	1.1364	0.0871	14
1.78	7.04	7.90	1	1.2044	0.1049	18
1.78	7.04	14.70	0	1.2008	0.0561	13
1.78	7.04	14.70	1	1.2969	0.1134	19
1.78	7.04	21.70	0	1.3872	0.1463	14
1.78	7.04	21.70	1	1.3725	0.1332	16
1.78	7.04	28.70	0	1.4467	0.1358	13
1.78	7.04	28.70	1	1.4076	0.0975	17
1.78	9.47	1.15	0	1.0928	0.0604	13
1.78	9.47	1.15	1	1.1358	0.0985	14
1.78	9.47	7.90	0	1.2164	0.0671	14
1.78	9.47	7.90	1	1.2809	0.1368	17
1.78	9.47	14.70	0	1.2832	0.1032	14
1.78	9.47	14.70	1	1.3139	0.1097	15
1.78	9.47	21.70	0	1.4264	0.1147	13
1.78	9.47	21.70	1	1.4347	0.1016	17
1.78	9.47	28.70	0	1.4962	0.1330	14

1.78	9.47	28.70	1	1.4533	0.0843	18
1.78	12.12	1.15	0	1.1399	0.1224	15
1.78	12.12	1.15	1	1.1489	0.1143	18
1.78	12.12	7.90	0	1.2097	0.1189	14
1.78	12.12	7.90	1	1.2551	0.1237	16
1.78	12.12	14.70	0	1.3082	0.0945	14
1.78	12.12	14.70	1	1.3544	0.1087	17
1.78	12.12	21.70	0	1.4660	0.1020	16
1.78	12.12	21.70	1	1.4589	0.0715	17
1.78	12.12	28.70	0	1.5580	0.1070	12
1.78	12.12	28.70	1	1.5439	0.1130	14
2.77	2.14	1.15	0	1.1286	0.0864	19
2.77	2.14	1.15	1	1.1740	0.1030	18
2.77	2.14	7.90	0	1.1446	0.1082	18
2.77	2.14	7.90	1	1.2519	0.1356	17
2.77	2.14	14.70	0	1.1947	0.1141	16
2.77	2.14	14.70	1	1.2458	0.1100	15
2.77	2.14	21.70	0	1.2366	0.0832	19
2.77	2.14	21.70	1	1.2826	0.0739	18
2.77	4.55	1.15	0	1.0743	0.0948	19
2.77	4.55	1.15	1	1.1444	0.0832	18
2.77	4.55	7.90	0	1.1060	0.0825	17
2.77	4.55	7.90	1	1.1803	0.0794	16
2.77	4.55	14.70	0	1.2059	0.1339	14
2.77	4.55	14.70	1	1.2426	0.1001	17
2.77	4.55	21.70	0	1.2415	0.0932	18
2.77	4.55	21.70	1	1.2956	0.0840	17
2.77	7.04	1.15	0	1.0887	0.1357	13
2.77	7.04	1.15	1	1.0684	0.0710	17
2.77	7.04	7.90	0	1.1550	0.0891	15
2.77	7.04	7.90	1	1.1735	0.0918	16
2.77	7.04	14.70	0	1.2227	0.0494	16
2.77	7.04	14.70	1	1.2850	0.0975	17
2.77	7.04	21.70	0	1.3152	0.0704	16
2.77	7.04	21.70	1	1.3370	0.1045	16
2.77	9.47	1.15	0	1.0825	0.0769	16
2.77	9.47	1.15	1	1.0732	0.0936	15
2.77	9.47	7.90	0	1.1636	0.1027	13
2.77	9.47	7.90	1	1.1643	0.0816	16
2.77	9.47	14.70	0	1.2549	0.0932	15
2.77	9.47	14.70	1	1.2994	0.1201	14
2.77	9.47	21.70	0	1.4033	0.1247	17
2.77	9.47	21.70	1	1.3672	0.0733	13
2.77	12.12	1.15	0	1.0746	0.1185	19
2.77	12.12	1.15	1	1.0993	0.0906	17
2.77	12.12	7.90	0	1.2218	0.0899	16
2.77	12.12	7.90	1	1.2717	0.1241	16
2.77	12.12	14.70	0	1.3823	0.1087	16
2.77	12.12	14.70	1	1.3537	0.0786	18
2.77	12.12	21.70	0	1.4593	0.1087	15
2.77	12.12	21.70	1	1.4222	0.0964	15
4.17	2.14	1.15	0	1.1386	0.1129	18
4.17	2.14	1.15	1	1.1865	0.1104	19
4.17	2.14	7.90	0	1.1506	0.1114	16
4.17	2.14	7.90	1	1.1770	0.1225	18
4.17	2.14	14.70	0	1.1784	0.1057	15
4.17	2.14	14.70	1	1.2362	0.0930	16
4.17	4.55	1.15	0	1.1371	0.0613	15
4.17	4.55	1.15	1	1.0790	0.1045	16
4.17	4.55	7.90	0	1.1141	0.1030	17
4.17	4.55	7.90	1	1.1386	0.1111	16
4.17	4.55	14.70	0	1.1820	0.1309	16
4.17	4.55	14.70	1	1.1954	0.1053	16
4.17	7.04	1.15	0	1.0657	0.1156	16
4.17	7.04	1.15	1	1.0486	0.1242	17
4.17	7.04	7.90	0	1.0868	0.1182	14
4.17	7.04	7.90	1	1.1852	0.1056	15
4.17	7.04	14.70	0	1.2740	0.0809	17
4.17	7.04	14.70	1	1.2641	0.1070	17
4.17	9.47	1.15	0	1.0384	0.0779	16
4.17	9.47	1.15	1	1.0384	0.0785	15
4.17	9.47	7.90	0	1.1732	0.0878	14
4.17	9.47	7.90	1	1.1764	0.1168	14
4.17	9.47	14.70	0	1.3334	0.1281	15
4.17	9.47	14.70	1	1.2955	0.1056	15
4.17	12.12	1.15	0	1.0670	0.1072	19
4.17	12.12	1.15	1	1.0686	0.1202	15
4.17	12.12	7.90	0	1.2105	0.1251	14
4.17	12.12	7.90	1	1.1665	0.0940	13
4.17	12.12	14.70	0	1.3402	0.0946	16
4.17	12.12	14.70	1	1.2870	0.0596	16